

IN THE CLAIMS

Please cancel claims 7-11, 14 and 17 and amend claims 1-6, 12, 13, 15 and 16 as follows:

1. (Currently Amended) A gas turbine combustor comprising ~~diffusive combustion nozzles which inject fuel and air into a combustion chamber and form a diffusive combustion flame, outer and inner walls which form an annular premixing flow passage and premixing nozzles which are disposed in the premixing flow passage and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into the combustion chamber, characterized in that~~

et ~~a plurality of the premixing nozzles are arranged in the premixing flow passage;~~

~~opening portions permitting air to flow in are provided at the outer wall so that the air flowed into the premixing flow passage forms swirling flow with respect to the premixing nozzles; and~~

~~the opening portions are disposed in circumferential direction and are provided one for every adjacent two remixing nozzles.~~ a combustion chamber, diffusive combustion nozzles which inject fuel into air in said combustion chamber to form a diffusive combustion flame, an annular premixing flow passage formed by outer and inner walls in said combustion chamber and premixing nozzles disposed in said premixing flow

passage for injecting fuel therein to mix with air to form a premixed gas, which flows out into said combustion chamber to form a premixing flame, characterized in that

a plurality of said premixing nozzles are mounted in spaced relationship in said premixing flow passage;

a plurality of spaced openings are formed in said outer wall through which air flows to mix with fuel from said premixing nozzles and form a swirling flow with respect to each of said premixing nozzles; and

said openings are disposed in a circumferential direction whereby one opening is provided for each two adjacent nozzles.

et
2. (Currently Amended) A gas turbine combustor comprising ~~diffusive combustion nozzles which inject fuel and air into a combustion chamber and form a diffusive combustion flame, outer and~~

~~inner walls which form an annular premixing flow passage and premixing nozzles which are disposed in the premixing flow passage and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into the combustion chamber, characterized in that~~

~~a plurality of the premixing nozzles are arranged in the premixing flow passage;~~

~~opening portions permitting air to flow in are provided at the outer wall so that the air flowed into the premixing flow passage forms swirling flow with respect to the premixing nozzles; and~~

~~the opening portions are disposed in circumferential direction and are provided one for every adjacent two remixing nozzles and the rotating directions of the swirling flows for the respective two premixing nozzles are caused to direct opposite directions each other according to claim 1 wherein said swirling flows for said two adjacent nozzles rotate in opposite directions.~~

3. (Currently Amended) A gas turbine combustor comprising:

A combustion chamber,

diffusive combustion nozzles which inject fuel ~~and to mix with air into a in said~~ combustion chamber and form a diffusive combustion flame;

an inner cylinder arranged outside the diffusive combustion nozzles;

a plurality of premixing nozzles which are arranged outside the inner cylinder in a circumferential direction and ~~form a premixing combustion flame by injecting inject fuel to mix air and form a premixed gas formed by premixing fuel and~~

air into which flows out into the combustion chamber to form a premixing flame;

and

means for forming ~~respective a swirling flows~~ flow of air and fuel ~~different rotating direction for the adjacent~~ around each of two adjacent premixing nozzles ~~in circumferential direction~~ whereby said flows rotate in opposite directions.

4. (Currently Amended) A gas turbine combustor according to claim 1, characterized in that each of ~~the opening portions~~ said openings is provided between the two adjacent ~~two~~ premixing nozzles ~~at the~~ in a position in a circumferential direction.

5. (Currently Amended) A gas turbine combustor according to claim 1, characterized in that each of ~~the opening portions~~ said openings is configured in such a manner that ~~the opening~~ a width of each opening in a circumferential direction varies along ~~the~~ an axial direction thereof.

6. (Currently Amended) (Currently Amended) A gas turbine combustor according to claim 5, characterized in that each of ~~the opening portions~~ said openings is configured in ~~nearly a triangular~~ a planform trapezoid shape in such a manner either that the opening broadens in ~~the~~ a main air

stream direction prior to air flowing into the ~~premixer~~
premixing flow passage or that the opening decreases in the
 main air stream direction prior to air flowing into the
~~premixer~~ premixing flow passage.

7-11. (Cancelled)

12. (Currently Amended) A gas turbine combustor ~~use~~
using a premixing device comprising a plurality of premixing
 nozzles which are arranged in a circumferential direction and
 form a premixing combustion flame by injecting premixed gas
 formed by premixing fuel and air into a combustion chamber,
 characterized in that one air flow inlet for every adjacent
two premixing nozzles is provided so that a swirling flow is
~~formed for the respective adjacent with respect to each of~~
~~said adjacent two premixing nozzles in the circumferential~~
~~direction.~~

13. (Currently Amended) A gas turbine combustor ~~use~~
using a premixing device comprising a plurality of premixing
 nozzles which are arranged in a circumferential direction and
 form a premixing combustion flame by injecting premixed gas
 formed by premixing fuel and air into a combustion chamber,
 characterized in that one air flow inlet for every adjacent
 two premixing nozzles is provided so that a swirling flows of
~~which rotating directions are opposite each other are formed~~

~~for the respective adjacent two premixing nozzles in the circumferential direction~~ flow is formed with respect to each of said two adjacent premixing nozzles and said swirling flows for said two adjacent nozzles rotate in opposite directions.

14. (Cancelled)

15. (Currently Amended) A premixing method for a gas turbine combustor comprising a plurality of premixing nozzles which are arranged in a circumferential direction and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into a combustion chamber, characterized in that ~~air is flown~~ flows in from air flow inlets, ~~each~~ with one of said air inlets being provided for every adjacent two premixing nozzles ~~in the circumferential direction~~, and a swirling flows ~~are~~ flow is formed around the ~~respective~~ each of said adjacent two premixing nozzles.

16. (Currently Amended) A premixing method for a gas turbine combustor comprising a plurality of premixing nozzles which are arranged in circumferential direction and form a premixing combustion flame by injecting premixed gas formed by premixing fuel and air into a combustion chamber, characterized in that ~~air is flown~~ flows in from air flow inlets, ~~each~~ with one of said inlets being provided for every adjacent two premixing nozzles, and a swirling flows ~~of which~~

~~rotating directions are opposite each other are~~ is formed
around ~~the respective~~ said adjacent two premixing nozzles, and
said swirling flows for said two adjacent nozzles rotate in
opposite directions.

17. (Cancelled)
